

ABSTRACT

The present invention provides a gas turbine vane with an airfoil cooling circuit incorporating multiple generally radially extending rows of pedestals generally bisected by a plurality of axially extending ribs with the ribs containing a plurality of recessed cavities. The recessed cavities are positioned immediately adjacent pedestals located closest to the ribs such that a cavity passageway is formed to pass sufficient cooling air between the rib and pedestal to increase overall heat transfer and reduce pressure loss of the cooling fluid.